

Special and General Relativity (PHZ 4601/5606) Fall 2017 Solutions

Set 4

10. Proper time under rotation.

The velocity v of the satellite is related to the angular frequency ω by $v = \omega r$. Therefore,

$$c^2 d\tau^2 = c^2 dt^2 - \omega^2 r^2 dt^2 \Rightarrow \frac{d\tau}{dt} = \sqrt{1 - \frac{\omega^2 r^2}{c^2}}.$$

For the gravitational acceleration a_g we have the equation

$$a_g = \frac{G m_e}{r_s^2} = \frac{v^2}{r_s},$$

where r_s is introduced for the radius of the stationary orbit. Eliminating v gives

$$\frac{G m_e}{r_s^2} = \omega^2 r_s = r_s \left(\frac{2\pi}{T} \right)^2 \Rightarrow r_s^3 = G m_e \left(\frac{T}{2\pi} \right)^2,$$

i.e., Kepler's 3rd law. For a stationary satellite, as well as for a point on the equator, we have $T = 24 \times 3600 [s]$ for the period. Crunching the numbers, the result for r_s is

$$r_s = \left[G m_e \left(\frac{T}{2\pi} \right)^2 \right]^{1/3} = 4.22 \times 10^7 [m],$$

which is about 6.6 times the earth radius.

From the first equation we find

$$\begin{aligned} \frac{d\tau_s}{d\tau_e} &= \frac{d\tau_s}{dt} \frac{dt}{d\tau_e} = \frac{\sqrt{1 - \omega^2 r_s^2/c^2}}{\sqrt{1 - \omega^2 r_e^2/c^2}} \\ &\approx \left(1 - \frac{1}{2} \frac{\omega^2 r_s^2}{c^2} \right) \left(1 + \frac{1}{2} \frac{\omega^2 r_e^2}{c^2} \right) \approx 1 - \frac{1}{2} \frac{\omega^2 r_s^2}{c^2} + \frac{1}{2} \frac{\omega^2 r_e^2}{c^2}. \end{aligned}$$

The numerical values are

$$\frac{1}{2} \frac{\omega^2 r_s^2}{c^2} = 5.249 \times 10^{-11} \quad \text{and} \quad \frac{1}{2} \frac{\omega^2 r_e^2}{c^2} = 1.194 \times 10^{-12}.$$

The combined, measurable, effect is therefore

$$1 - \frac{d\tau_s}{d\tau_e} = \frac{1}{2} \frac{\omega^2 r_s^2}{c^2} - \frac{1}{2} \frac{\omega^2 r_e^2}{c^2} = 5.130 \times 10^{-11},$$

i.e., the clock on the satellite ticks slower.

From gravity we obtain

$$1 - \frac{d\tau_s}{d\tau_e} = -\frac{G m_e}{c^2} \left(\frac{1}{r_s} - \frac{1}{r_e} \right) = -5.911 \times 10^{-10},$$

which is about ten times larger than the SR effect.